### SHIRT BUTTON-SIZED MICROTURBOJET ENGINES

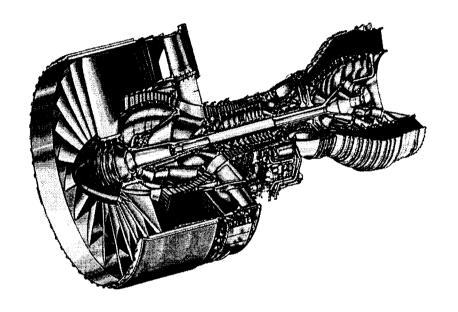
# **Presented by**

Professor Alan H. Epstein Massachusetts Institute of Technology

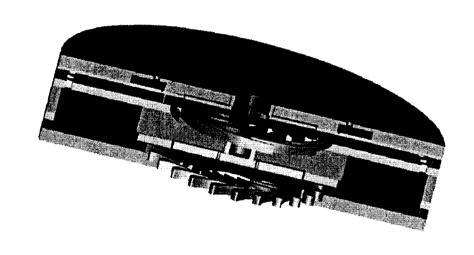
December 1998

## "MACRO" vs. "MICRO" GAS TURBINES

#### "MACRO"



### "MICRO"



10,000 parts

Inlet dia = 2.8 m

Airflow = 1100 kg/sec

Weight = 10 tons

Thrust = 400,000 Nt (50

MW)

**Price** ~ \$30/Nt

2 parts

Inlet dia = 2 mm

Airflow = 0.4 g/sec

Weight = 1 gram

Thrust = 0.1-0.2 Nt (10-30)

watts)

### HIGH POWER DENSITY THERMODYNAMIC CYCLES

- Physical Requirements are Invariant With Size -

- High peak cycle temperatures (1200 ~ 1700°K)
  - High temperature materials
- High peripheral speeds (400 600 m/s), thus [Fluid & electric power density  $\infty$  (Tip speed)<sup>2</sup>  $\infty$  Stress]
  - Highly stressed rotating parts (100's MPa)
- Low friction bearings
- Reasonable component efficiencies
  - Close tolerances (1 μm)

# **μENGINES ARE NOT SCALED-DOWN BIG ENGINES**

<u>Physics Intrinsic to Small Devices</u> <u>Current Technology Limits</u>

Fluid viscous effects up 2–D "extruded" shapes

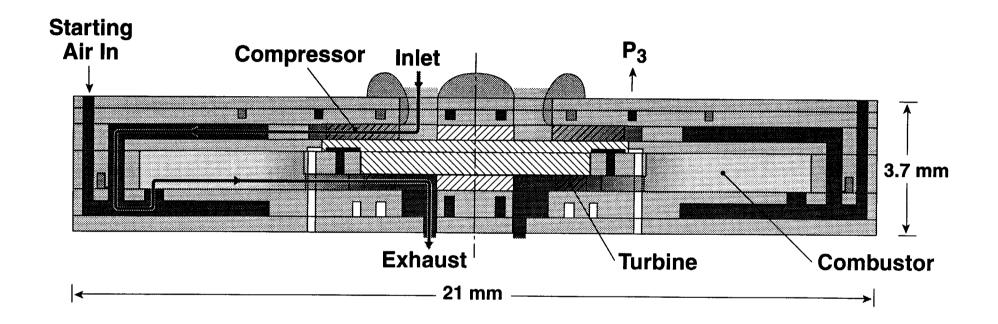
Surface area to volume high Etching depth & aspect ratio

Short heat conduction paths Number of layers (~10+)

Chemical reaction times const. Most fab tech applies to Si

Many materials are stronger Assembly, packaging, integration

# H<sub>2</sub> DEMO ENGINE - Silicon, Cooled Turbine -



Thrust = 11 g

Fuel burn = 16 g/hr

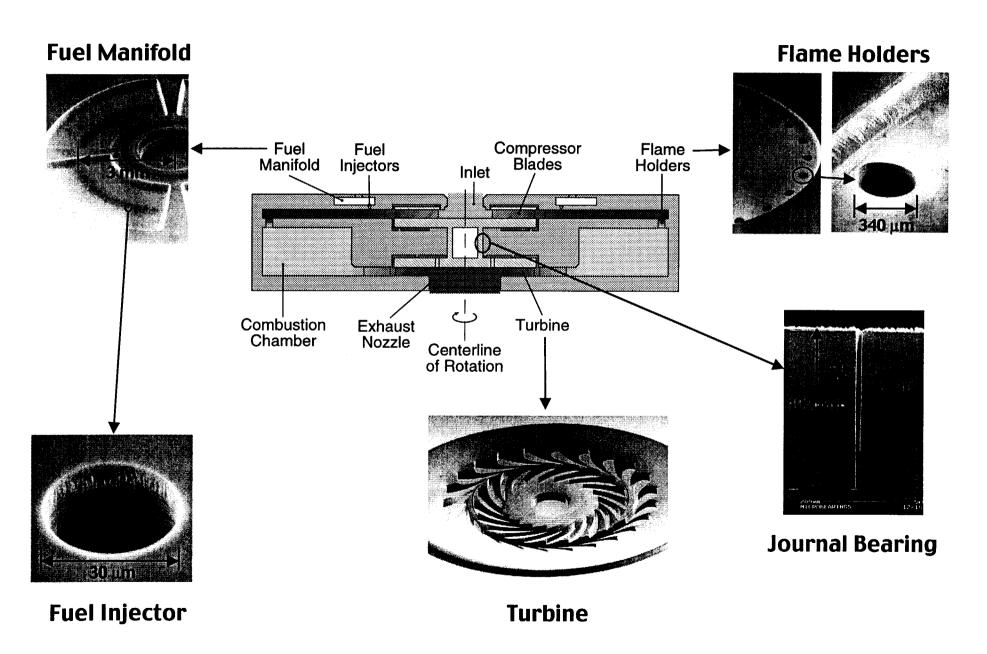
**Engine weight = 1 gram** 

Turbine inlet temp = 1600°K (2421°F)

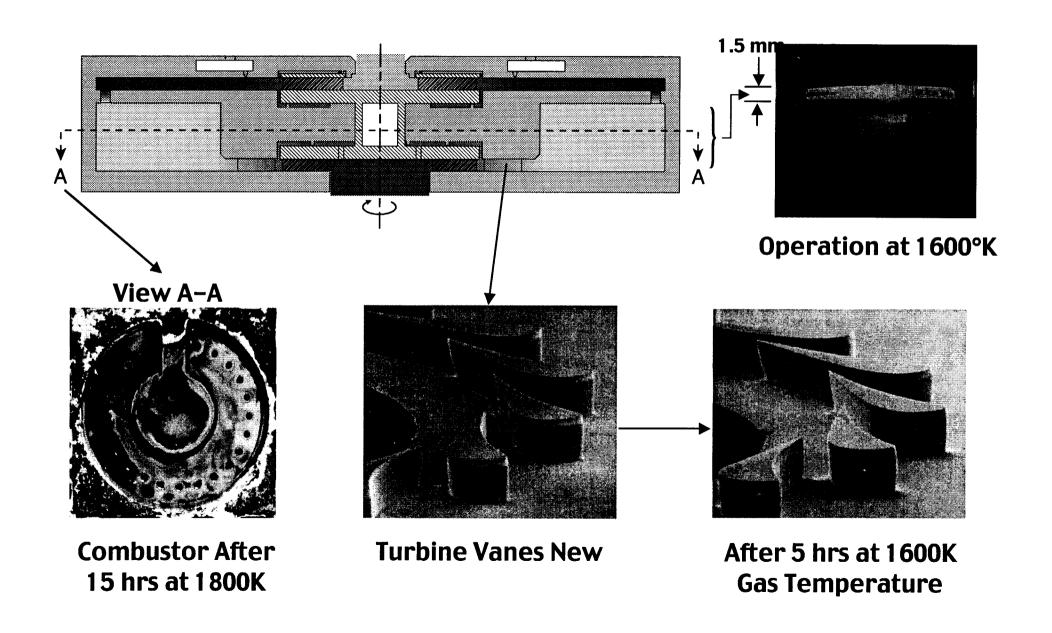
Rotor speed =  $1.2 \times 10^6$  RPM

Exhaust gas temp = 970°C

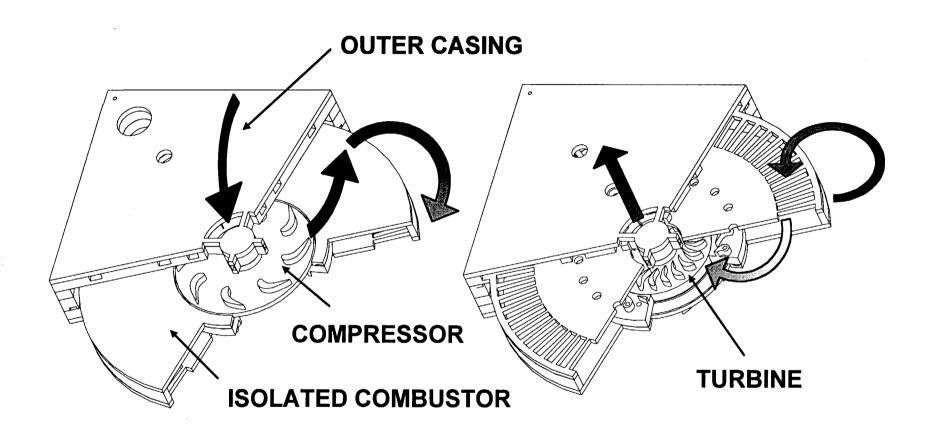
# **MICROENGINE FABRICATION DETAILS**



### MICROCOMBUSTOR MECHANICAL INTEGRITY



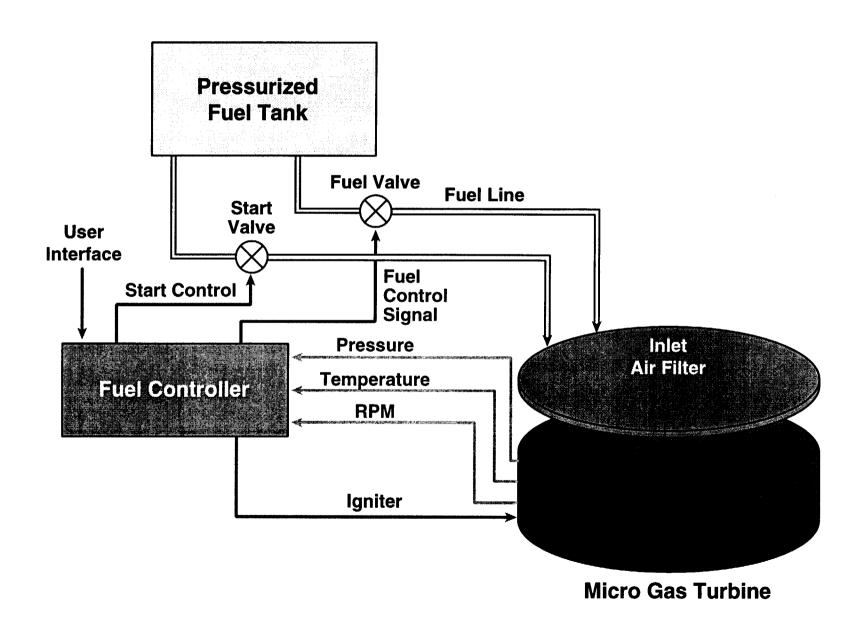
## **3D CUTAWAY OF DEMO ENGINE**



**LOOKING AFT** 

**LOOKING FORWARD** 

# **AUXILIARY SYSTEMS AND COMPONENTS**



### MICRO GAS TURBINE ENGINE PROGRAM PLAN

